

# Flow and Noise Associated with The Interaction of A Square Cylinder with A Downstream Flat Plate



Mohamed Sukri Mat Ali  
School of Mechanical Engineering  
University of Adelaide

A thesis submitted for the degree of  
*Doctor of Philosophy*

2011

*I would like to dedicate this thesis to my late father...*  
*Mat Ali Samsudin*

## Acknowledgements

The completion of this thesis was supported financially by the Ministry of Higher Education of Malaysia through the Universiti Teknologi Malaysia. The use of supercomputer facilities provided by eReserachSA to expedite the research work is also appreciated greatly.

I have been indebted in the preparation of this thesis to my supervisors, Dr. Con J. Doolan of University of Adelaide, Dr. Vincent Wheatley of University of Queensland and Dr. Laura Brooks of University of Adelaide, whose patience and kindness, as well as their academic experience, have been invaluable to me.

Thanks to my family, especially to my mother, for their emotional and moral support in the completion of this thesis.

I am extremely grateful to Mr. Billy Constantine for his assistance on the technical difficulties related to my research. I also thank the staff members in the front office of the School of Mechanical Engineering and my friends for their kindness and various types of assistance.

## Declarations

This work contains no material which has been accepted for the award of any other degree or diploma in any university or other tertiary institution to MOHAMED SUKRI MAT ALI and, to the best of my knowledge and belief, contains no material previously published or written by another person, except where due reference has been made in the text.

I give consent to this copy of my thesis when deposited in the University Library, being made available for loan and photocopying, subject to the provisions of the Copyright Act 1968. The author acknowledges that copyright of published works contained within this thesis resides with the copyright holder(s) of those works.

I also give permission for the digital version of my thesis to be made available on the web, via the University's digital research repository, the Library catalogue, the Australasian Digital Theses Program (ADTP) and also through web search engines, unless permission has been granted by the University to restrict access for a period of time.

Date: 15<sup>th</sup> November 2011

---

Mohamed Sukri Mat Ali

## Abstract

Bluff bodies are commonly found in many engineering applications, such as aircraft landing gear, pantograph systems of high speed trains and the rear view mirrors of passenger cars. In the event of bluff body interaction with flow, various flow induced phenomena may occur, such as vibration, unsteady aerodynamic loading, high pressure drag and flow induced noise. These phenomena can be altered by locating a secondary body in the wake of, and close to the bluff body. This alteration is due to the modification of the bluff body wake structure by the secondary body.

The main objective of this study is to investigate the possibility of using a downstream flat plate for passive bluff body wake control. The success of this mechanism is evaluated by considering the reduction in unsteady aerodynamic loading together with the reduction in sound pressure level at an observer location. The observed new flow structures are explained using the results from a wide range of wake analyses to obtain a clear understanding of the physical phenomena of the wake-plate interaction.

The test case under investigation consists of a square cylinder as the primary bluff body and a flat plate as the secondary body. The plate length and its position downstream of the square cylinder are varied systematically. This allows the optimal configuration of the plate for the least aerodynamic loading and radiated sound to be determined. The flow is simulated using two-dimensional Direct Numerical Simulation (DNS) at a Reynolds number of 150, based on the side length of the cylinder of  $D$ . For the acoustic analysis, Curle's solution of Lighthill's acoustic analogy is used for a Mach number of 0.2 and the

sound pressure levels are calculated for an observer location of  $80D$  directly above the cylinder. The results of the study are presented in the form of a chapter describing the mesh verification and validation studies and a collection of four published or submitted journal articles on the remaining results.

The effect of an attached downstream flat plate on the wake of a square cylinder is first investigated for plate lengths ( $L$ ) of length  $0 \leq L \leq 6D$ . The plate, which is called a splitter plate, is attached to the downstream edge of the cylinder. It is found that the splitter plate interacts strongly with the near wake of the cylinder and that the length of the plate significantly affects the flow structure. The behaviour of the flow can be grouped into three regimes. For short plate lengths  $0 < L \lesssim D$ , the free shear layers are convected further downstream before rolling up into vortices as the plate length is increased. In this regime, the root mean square lift coefficient ( $C_{L_{\text{rms}}}$ ) and the mean drag ( $C_{D_{\text{mean}}}$ ) decrease with increasing plate length. For intermediate plate lengths  $1.25D \lesssim L \lesssim 4.75D$ , a secondary vortex is clearly visible around the trailing edge of the splitter plate and the shear layers begin to roll up closer to the trailing edge. A continued increase in  $C_{L_{\text{rms}}}$  is observed when the plate length is increased, but a reduction in  $C_{D_{\text{mean}}}$  still occurs, although at a lesser rate than the first regime. For long plate lengths  $L \gtrsim 5D$ , a regime is observed in which the free shear layers reattach to the splitter plate. Consequently, a sudden jump in  $C_{L_{\text{rms}}}$  is observed. The splitter plate study also proposes the minimum wake half-width as the length scale for a possible universal Strouhal number, which is found to be valid for  $0 \leq L \leq 4D$ .

The effect of the splitter plate on sound generation is also numerically investigated. When the length of the splitter plate is varied from  $L = 0.5D$  to  $6D$  the results can also be grouped into three distinct regimes. For the first regime ( $L \lesssim D$ ), the sound pressure levels decrease with increasing plate length. An overall sound pressure level

reduction of 3 dB is obtained at the observer location when the length of the plate is  $D$ . For the second regime ( $1.25D \lesssim L \lesssim 4.75D$ ), the sound levels increase with increasing plate length. For the third regime ( $L \gtrsim 5D$ ), the sound pressure levels decrease as the length of the plate increases but the levels are higher than for the other regimes. Results also show that the lift fluctuation is the dominant sound source. These acoustic results are explained in terms of the fluid mechanics occurring in the near wake of the cylinder.

In the second part of the study, the sensitivity of the square cylinder wake structure to a detached plate is investigated by varying the gap distance ( $G$ ) between the cylinder and plate along the wake centerline in the range  $0 \leq G \leq 7D$ . The length of the plate is kept constant and equal to the side length of the square cylinder. A critical gap distance, which separates the two flow regimes, is observed to occur at  $G_c \approx 2.3D$ . Regime I is characterised by vortex formation occurring downstream of the gap while for regime II, formation occurs within the gap. Reductions in  $C_{L_{\text{rms}}}$  and  $C_{D_{\text{mean}}}$  can only be obtained in regime I, where the optimal gap occurs at the critical gap. When the gap distance is increased beyond its critical point (regime II), a significant increase for both parameters is observed. This is due to the re-establishment of the von Kármán vortex inside the gap.

The sound generation for the detached plate case is numerically investigated. The study found that the change in the sound pressure level with gap distance can be grouped according to the flow regimes identified earlier. An overall sound pressure reduction to the freestream can only be obtained in regime I ( $0 \leq G \leq 2.3D$ ), where a 2.9 dB reduction in sound pressure level is obtained when there is no gap between the two bodies. In contrast, the sound pressure level increases by at least 8.0 dB when the plate is in regime II ( $2.4D \leq G \leq 7D$ ).

Using the information from the flow simulation, the plate length is reduced in an effort to achieve sound cancellation in the far-field. This can be achieved when the noise generated by the cylinder and the plate

are equal in magnitude but out of phase. Using a linear estimation method, the plate length and gap distance required to obtain sound cancellation are found to be  $L = 0.26D$  and  $G = 5.6D$ , respectively. Acoustic analysis of the modified configuration shows a sound pressure reduction of 6.3 dB at the observer location when compared with the single square cylinder. A detailed flow-visualisation shows that the maximum sound pressure reduction attainable is limited by a non-linear unsteady stall process on the plate.

# Contents

<b>Declarations</b>	<b>iii</b>
<b>Abstract</b>	<b>iv</b>
<b>Contents</b>	<b>viii</b>
<b>List of Figures</b>	<b>xii</b>
<b>List of Tables</b>	<b>xv</b>
<b>1 Introduction</b>	<b>1</b>
1.1 Research background . . . . .	1
1.2 Aims and significance of the research . . . . .	3
1.3 Problem geometry . . . . .	4
1.4 Thesis structure . . . . .	5
References . . . . .	7
<b>2 Background and Literature Review</b>	<b>10</b>
2.1 Overview of the flow over a bluff body . . . . .	10
2.2 Flow over a bluff body with passive wake control . . . . .	14
2.2.1 Flow over a bluff body with a splitter plate . . . . .	15
2.2.2 Flow over a bluff body with a detached flat plate . . . . .	22
2.2.2.1 Pre-vortex formation regime . . . . .	25
2.2.2.2 Post-vortex formation regime . . . . .	25
2.3 Sound radiated from bluff bodies . . . . .	27
2.3.1 Passive sound control using a splitter plate . . . . .	29

2.3.2	Passive sound control using a downstream detached flat plate	30
2.4	Research gaps . . . . .	32
2.5	Summary of the journal articles . . . . .	34
2.5.1	Low Reynolds number flow over a square cylinder with a splitter plate (Paper I [99], Chapter 5) . . . . .	34
2.5.2	The sound generated by a square cylinder with a splitter plate a low Reynolds number (Paper II [100], Chapter 6) . . . . .	36
2.5.3	Low Reynolds number flow over a square cylinder with a detached flat plate (Paper III [98], Chapter 7) . . . . .	37
2.5.4	Aeolian tones generated by a square cylinder with a detached flat plate (Paper IV [101], Chapter 8) . . . . .	38
2.6	How the current findings can be applied to the real applications . . . . .	40
	References . . . . .	42
<b>3</b>	<b>Solution Methodologies</b>	<b>53</b>
3.1	Flow simulations . . . . .	53
3.1.1	Discretisation of the governing equations . . . . .	54
3.1.1.1	Discretisation of the convective term . . . . .	55
3.1.1.2	Discretisation of the diffusion term . . . . .	59
3.1.1.3	Temporal discretisation . . . . .	60
3.1.2	Algebraic equations for the Navier-Stokes equations . . . . .	62
3.2	Acoustic calculation . . . . .	65
3.3	The computing machine . . . . .	70
	References . . . . .	71
<b>4</b>	<b>Grid Refinement Study</b>	<b>75</b>
4.1	Introduction . . . . .	76
4.2	Test case and solution methodology . . . . .	78
4.2.1	Governing equations . . . . .	79
4.2.2	Solution methodology . . . . .	79
4.2.3	Flow initialisation, data capture and boundary conditions . . . . .	81
4.3	Grid refinement stage I . . . . .	81
4.3.1	Flow visualisation . . . . .	85

4.3.2	Comparisons of Strouhal number ( $St$ ), root mean square lift coefficient ( $C_{L_{rms}}$ ) and mean drag coefficient ( $C_{D_{mean}}$ ) . . . . .	87
4.3.3	Findings from grid refinement stage I . . . . .	88
4.4	Grid refinement stage II . . . . .	88
4.4.1	Thwaites' method . . . . .	89
4.4.2	Results from applying Thwaites' method . . . . .	92
4.4.3	The influence of the outlet boundary condition . . . . .	97
4.4.4	The influence of temporal discretisation schemes . . . . .	99
4.4.5	The influence of convective discretisation schemes . . . . .	101
4.4.6	Grid convergence study using Richardson extrapolation . . . . .	101
4.4.6.1	A generalization of Richardson extrapolation . . . . .	102
4.4.6.2	Level of grid convergence . . . . .	103
4.5	Comparison between incompressible and compressible solutions for case E . . . . .	112
4.6	Comparison of numerical data from case E with previous studies . . . . .	113
4.7	Chapter summary . . . . .	113
	References . . . . .	115
<b>5</b>	<b>Low Reynolds Number Flow Over A Square Cylinder with A Splitter Plate</b>	<b>118</b>
<b>6</b>	<b>The Sound Generated by A Square Cylinder with A Splitter Plate at Low Reynolds Number</b>	<b>133</b>
<b>7</b>	<b>Low Reynolds Number Flow over A Square Cylinder with A Detached Plate</b>	<b>152</b>
<b>8</b>	<b>Aeolian Tones Generated by A Square Cylinder with A Detached Flat Plate</b>	<b>191</b>
<b>9</b>	<b>Conclusions and Recommendations for Future Work</b>	<b>224</b>
9.1	Conclusions . . . . .	224
9.2	Recommendations for future work . . . . .	228
9.2.1	The sound cancellation was limited by the plate stall phenomenon . . . . .	228

9.2.2	The spanwise flow structure becomes important at high Reynolds number . . . . .	228
9.2.3	The effect of plate thickness . . . . .	229
9.2.4	The effect of cylinder chord length . . . . .	229
9.2.5	The effect of cross-stream position of the flat plate . . . . .	229
<b>Appendix A</b>		<b>230</b>
<b>Appendix B</b>		<b>237</b>
<b>Appendix C</b>		<b>246</b>

# List of Figures

1.1	A square cylinder with a flat plate immersed in a free stream velocity of $U_\infty$ .	5
2.1	Variation of Strouhal numbers with Reynolds numbers. $\circ$ : Experiments of Williamson [3] for a circular cylinder, $\square$ : experiments of Luo et al.[14] for a square cylinder and $\nabla$ : experiments of Norberg (data taken from ref.[4]) for a square cylinder. Solid lines are from the empirical relation of Eq. (2.1.2).	12
2.2	The effects of splitter plate length on Strouhal number ( $St$ ) from selected previous studies. Kwon and Choi [39] for a circular cylinder at low Reynolds numbers: $Re = 80$ ; $\circ$ , $Re = 100$ ; $\square$ , $Re = 120$ ; $\diamond$ , $Re = 140$ ; $\nabla$ , $Re = 160$ ; $\triangle$ . You et al. [40] for a circular cylinder at low Reynolds numbers: $Re = 100$ ; $- + -$ , $Re = 160$ ; $- \times -$ . Nakamura [41] for a circular cylinder at high Reynolds numbers: $Re = 1000$ ; $- \circ -$ , $Re = 5300$ ; $-\square-$ , and a semi-circular cylinder: $Re = 1000$ ; $\triangleleft$ Anderson and Szewczyk [42] for a circular cylinder at high Reynolds number: $Re = 4600$ ; $-\diamond-$ .	16
2.3	The effects of splitter plate length on base pressure ( $-C_{pb}$ ) from selected previous studies. $-C_{pb}$ : Apelt and West [38] for a circular cylinder ( $Re \sim 10^4$ ); $\circ$ and for a '†' shape cylinder; $\diamond$ . Bearman [30] for a rectangular cylinder with an ogive leading edge ( $Re = 2.45 \times 10^5$ ); $\triangleleft$ .	19
2.4	Two regimes for the case of a bluff body with a detached flat plate. $G_{pre} < G_{post}$ .	23

2.5	Variations of Strouhal number ( $St$ ) with gap distance from selected previous studies. Roshko [28] for a circular cylinder at $Re = 1.45 \times 10^4$ ; $\circ$ . Ozono [58] for a circular cylinder and a normal plate at $Re = 2.5 \times 10^4$ ; $\triangleright$ for a circular cylinder and $\diamond$ for normal plate. Hwang et al. [59] for a circular cylinder at $Re = 160$ ; $\square$ . The broken lines indicate a transition from the pre-vortex formation regime into the post-vortex formation regime. . . . .	24
3.1	Small fluid element fixed in space . . . . .	55
3.2	Point interpolation and annotations shown in one-dimensional form (i.e., $x$ -direction only). . . . .	56
3.3	Control volume of an orthogonal grid system. $P$ and $E$ are the cell centres of control volumes with sharing face. . . . .	59
3.4	Approximation of time derivatives. . . . .	61
3.5	PISO algorithm . . . . .	64
3.6	Sound radiated from the unsteady flow field into the acoustic field (i.e., far-field). . . . .	66
4.1	Sketch of the test case and annotations. . . . .	78
4.2	Sketch of flow phenomena on flow around a square cylinder. . . . .	82
4.3	Grid structure for case A. Fig. 4.3(a) complete grid and Fig. 4.3(b) grid in the region around the square cylinder. . . . .	83
4.4	Grid structure for case B. Fig. 4.4(a) complete grid and Fig. 4.4(b) grid in the region around the square cylinder. . . . .	84
4.5	Instantaneous spanwise vorticity contours coloured by rotation direction (blue: clockwise and green: anticlockwise), 40 equally spaced over the range $-4 \leq \frac{\Omega D}{U_\infty} \leq 4$ . Fig. 4.5(a) is for case A and Fig. 4.5(b) is for case B. . . . .	86
4.6	Grid domain for grid refinement study II (not to scale). $L/D$ is the length of the grid regions inside the computational domain ( $D$ is the side length of the square cylinder). . . . .	90
4.7	A sketch of boundary layer profile for plane stagnation-point flow. . . . .	91
4.8	Edge velocity ( $u_e$ ) profile of potential flow along the half upstream edge of a square cylinder. . . . .	93

4.9	Boundary layer thicknesses distribution: Top; Boundary layer thickness ( $\delta$ ), Middle; Displacement thickness ( $\delta^*$ ) and Bottom; Momentum thickness ( $\theta$ ). . . . .	94
4.10	Boundary layer thickness comparison between Thwaites' method with case A and case B. Boundary layer thickness from the finest grid (case E) that is used in the second stage of grid refinement study is also plotted for a comparison. . . . .	95
4.11	Grid structure for case E. . . . .	98
4.12	Comparison of instantaneous streamwise vorticity contours colored by rotation direction (blue: clockwise and green: anticlockwise), 40 equally spaced over the range $-4 \leq \frac{\Omega D}{U_\infty} \leq 4$ . Fig. 4.12(a); DNS with constant pressure and zero gradient (case C) and Fig. 4.12(b); DNS with convective outlet (case F). . . . .	100
4.13	$C_{L_{rms}}$ comparison between DNS results of the three grid solutions and Richardson Extrapolation estimation. . . . .	106
4.14	Error estimation of $C_{L_{rms}}$ relative to Richardson Extrapolation. . . . .	106
4.15	$C_{D_{mean}}$ comparison between DNS results of the three grid solutions and Richardson Extrapolation estimation. . . . .	107
4.16	Error estimation of $C_{D_{mean}}$ relative to Richardson Extrapolation. . . . .	108
4.17	$St$ comparison between DNS results of the three grid solutions and Richardson Extrapolation estimation. . . . .	108
4.18	Error estimation of $St$ relative to Richardson Extrapolation. . . . .	109
4.19	Boundary layer thicknesses at the leading edge of the square cylinder for various grid solutions. . . . .	110
4.20	Magnitude different between simulation results and Richardson Extrapolation estimation. . . . .	111
4.21	Percent of RMS error for boundary layer thickness along the leading edge. The order-of-accuracy, $p = 2$ and $E1 = 0.278\%$ , $E2 = 0.773\%$ and $E3 = 1.695\%$ . . . . .	111
4.22	Wake profiles (i.e., mean streamwise velocity) for incompressible (solid lines) and compressible (broken lines) solutions for case E. . . . .	112

# List of Tables

1.1	Problem geometry configurations according to parameters modified.	6
2.1	Flow Regimes of disturbed cylinder wake for bluff bodies with a splitter plate from selected previous investigations . . . . .	22
3.1	Corvus performances for various number of cores used in the DNS of a single square cylinder. . . . .	70
4.1	Grid parameters for cases A and B. $N$ is the number of cells, $X$ and $Y$ are the distances of the computational boundary from the cylinder in the streamwise and cross-stream directions, respectively, $\eta$ is the cell size near the cylinder wall and $g$ is the grid stretching ratio. Subscripts $x$ and $y$ are streamwise and cross-stream directions, respectively, $u$ , $d$ , $tb$ and $c$ are upstream, downstream, top-bottom and wake centerline regions, respectively. . . . .	80
4.2	Comparison of the current DNS results with previous DNS investigations. . . . .	87
4.3	Grid parameters for cases C, D and E. Subscripts $u$ , $d$ and $tb$ represent upstream, downstream and top and bottom location, respectively, and subscripts (1), (2) and (3) represent region 1, 2 and 3 of the computational domain, respectively. The total number of nodes is $N_x \times N_y$ , the number of nodes along the cylinder edges is $G_{edge}$ , $g$ is cell size ratio between the adjacent cell and the cell area is $\Delta$ . . . . .	96

4.4	Comparison of global results between DNS with a conventional outlet boundary condition, i.e., fixed ambient pressure and fixed zero velocity gradient, (case C) and DNS with a convective outlet boundary condition (case F). . . . .	99
4.5	Comparison of global results between DNS with $2^{nd}$ -order accurate (case D) and $1^{st}$ -order accurate (case G) temporal discretisation. . .	101
4.6	Comparison of global results between DNS with second order up-wind scheme (case E) and third order QUICK scheme (case H) convective discretisation scheme. . . . .	101
4.7	Parameters used and results from the three grid solutions (C, D and E). . . . .	104
4.8	Order of accuracy and Grid Convergence Index for three integration variables. Subscripts 3, 2 and 1 represent case C, D and E, respectively. . . . .	104
4.9	Richardson Extrapolation ( <i>RE</i> ) estimation data and errors. . . .	106
4.10	Global parameters for grid refinement study of boundary layer thickness. The chosen order of accuracy $p^{th}$ is 2, as the calculated value is exceeded the order of accuracy of the discretisation scheme. . . . .	109
4.11	Comparison of global results between incompressible and compressible solutions for case E. . . . .	113
4.12	Comparison of current DNS with previous studies listed by Doolan [4]. . . . .	113